U S Department of Transportation
Federal Aviation Administration

MAJOR REPAIR AND ALTERATION
(Airframe, Powerplant, Propeller, or Appliance)

Form Approved OMB No. 2120-0020 11/30/2007	Electronic Tracking Number
For FAA Use Only	

INSTRUCTIONS: Print or type all entries. See FAR 43.9, FAR 43 Appendix B, and AC 43.9-1 (or subsequent revision thereof) for instructions and disposition of this form. This report is required by law (49 U.S.C. 1421). Failure to report can result in a civil penalty not to exceed $1,000 for each such violation (Section 901 Federal Aviation Act 1958)

1. Aircraft	Nationality and Registration Mark: N777ND	Serial No.: 1258	
	Make: CIRRUS	Model: SR20	Series:
2. Owner	Name *(As shown on registration certificate)*: DAC'S AVIATION LLC	Address *(As shown on registration certificate)*: Address PO BOX 99 City ABERDEEN State SD Zip 57402 Country USA	

3. For FAA Use Only

4. Type		5. Unit Identification			
Repair	Alteration	Unit	Make	Model	Serial Number
☐	☒	AIRFRAME		*(As described in Item 1 above)*	
☐	☐	POWERPLANT			
☐	☐	PROPELLER			
☐	☐	APPLIANCE	Type:		
			Manufacturer:		

6. Conformity Statement

A. Agency's Name and Address	B. Kind of Agency			
Name: Quest Aviation Inc.	☐	U.S. Certificated Mechanic	☐	Manufacturer
Address: 4440 E. HWY 12	☐	Foreign Certificated Mechanic	C. Certificate No.	
City: ABERDEEN State: SD	☒	Certificated Repair Station	OW5R108N	
Zip: 57401 Country: USA	☐	Certificated Maintenance Organization		

D. I certify that the repair and/or alteration made to the unit(s) identified in item 5 above and described on the reverse or attachments hereto have been made in accordance with the requirements of Part 43 of the U.S. Federal Aviation Regulations and that the information furnished herein is true and correct to the best of my knowledge.

Extended range fuel per 14 CFR Part 43 App. B ☐	Signature/Date of Authorized Individual: 5-14-2009 BLAKE BRAUN

7. Approval for Return to Service

Pursuant to the authority given persons specified below, the unit identified in item 5 was inspected in the manner prescribed by the Administrator of the Federal Aviation Administration and is ☒ APPROVED ☐ REJECTED

BY								
		FAA Flt Standards Inspector		Manufacturer		Maintenance Organization		Person Approved by Canadian Department of Transport
		FAA Designee	X	Repair Station		Inspection Authorization		Other *(Specify)*

Certificate or Designation No.: OW5R108N	Signature/Date of Authorized Individual: 5-14-2009 MARK LEHRKAMP

FAA Form 337 (10-06)

NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished

(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

N777ND — Nationality and Registration Mark

5-14-2009 — Date

Installed J.P. Instruments EGT-701 with the fuel flow option STC SA2586NM

Removed the Cirrus optional EGT/CHT indicator P/N EG2850-08033 S/N 304

Installation made in accordance with JPI's FAA approved installation manual #103 revision 3 dated 6-17-1999 and report #503 Revision B dated 3/14/1997

Supplement EGT-710 Rev. B inserted in the POH

Weight and balance supplement completed

Continued airworthiness instructions are contained in the JPI installation manuals.

END

Quest
AVIATION, INC.

EQUIPMENT CHANGE - WEIGHT & BALANCE

REG. NO.	MODEL		Serial No.
777ND	SR20		1258

Items: (Description / P/N / S/N)	Weight Pounds	Arm Inches	Moments Inch/Pounds
Previous Aircraft Empty Weight:	2135.57	141.33	301827.72
REMOVED: OPTIONAL EGT/CHT	-0.6	119	-71.4
			0
INSTALLED: JPI EGT-701 SCANNER	0.9	119	107.1
EGT/CHT PROBES	1.2	74	88.8
HARNESS	0.7	100	70
			0
			0
			0
			0
			0
			0
			0
			0
			0
			0
			0
Totals	2137.77		302022.22

A. Old Empty Weight	2135.57	Pounds
B. Old Empty CG	141.33	Inches
C. Old Empty Weight CG Moment	301827.7	Inch/Pounds
D. Max Gross Weight	3000	Pounds
E. Old Useful Load	864.43	Pounds

A. New Empty Weight	2137.77	Pounds
B. New Empty CG	141.2791	Inches
C. New Empty Weight CG Moment	302022.2	Inch/Pounds
D. Max Gross Weight	3000	Pounds
E. New Useful Load	862.23	Pounds

This new weight & balance information superseads all previous weight and balance data.
For aircraft loading, see instructions in Weight & Balance Section of Aircraft Flight Manual.

FAA Form 337 Completed?	YES		
Equipment List Amended?	YES		
Mark Lehrkamp		**Date:**	5/14/2009
CRS OW5R108N			

U S Department of Transportation
Federal Aviation Administration

MAJOR REPAIR AND ALTERATION
(Airframe, Powerplant, Propeller, or Appliance)

Form Approved OMB No 2120-0020 11/30/2007

Electronic Tracking Number

For FAA Use Only

INSTRUCTIONS. Print or type all entries. See FAR 43.9, FAR 43 Appendix B, and AC 43.9-1 (or subsequent revision thereof) for instructions and disposition of this form. This report is required by law (49 U.S.C. 1421). Failure to report can result in a civil penalty not to exceed $1,000 for each such violation (Section 901 Federal Aviation Act 1958)

1. Aircraft	Nationality and Registration Mark: USA N777ND	Serial No: 1258	
	Make: Cirrus	Model: SR20	Series:
2. Owner	Name *(As shown on registration certificate)*: DACS Aviation LLC	Address *(As shown on registration certificate)*: Address PO Box 99; City Aberdeen; State SD; Zip 57402; Country USA	

3. For FAA Use Only

The data identified herein complies with applicable airworthiness requirements and is approved only for the above described aircraft subject to conformity inspection by a person authorized in CFR 14 part 43.7

Mark C. Goodwin — *Date* — Rapid City Flight Standards District Office GL-27

4. Type		5. Unit Identification			
Repair	Alteration	Unit	Make	Model	Serial Number
☒	☐	AIRFRAME		*(As described in item 1 above)*	
☐	☐	POWERPLANT			
☐	☐	PROPELLER			
☐	☐	APPLIANCE	Type / Manufacturer		

6. Conformity Statement

A. Agency's Name and Address: Name Great Planes Avionics Services, Inc.; Address 47008 Great Planes Pl.; City Tea; State SD; Zip 57064; Country USA

B. Kind of Agency

☐	U.S. Certificated Mechanic	☐	Manufacturer
☐	Foreign Certificated Mechanic	C. Certificate No	
☒	Certificated Repair Station	GUJR842X	
☐	Certificated Maintenance Organization		

D. I certify that the repair and/or alteration made to the unit(s) identified in item 5 above and described on the reverse or attachments hereto have been made in accordance with the requirements of Part 43 of the U.S. Federal Aviation Regulations and that the information furnished herein is true and correct to the best of my knowledge.

Extended range fuel per 14 CFR Part 43 App. B ☐

Signature/Date of Authorized Individual: Donald J. Hayes 3-13-08

7. Approval for Return to Service

Pursuant to the authority given persons specified below, the unit identified in item 5 was inspected in the manner prescribed by the Administrator of the Federal Aviation Administration and is ☒ APPROVED ☐ REJECTED

BY		FAA Flt Standards Inspector		Manufacturer		Maintenance Oranaization	Person Approved by Canadian Department of Transport
		FAA Designee	X	Repair Station		Inspection Authorization	Other *(Specify)*

Certificate or Designation No. GUJR842X

Signature/Date of Authorized Individual: Donald J. Hayes 3-13-08

México S.
C.P. 5409

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NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished

(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

USA N777ND	3/13/08
Nationality and Registration Mark	Date

N777ND

Validated that the previous installation of GNS 430 was installed IAW Garmin instructions during manufacturing of aircraft. Verified this aircraft and all interfaced equipment are covered under the STC AML. Unit was removed and upgraded to the GNS 430W unit. The existing locations of the unit was determined to meet the field-of-view requirement with annunciation provided by Sandel 3308 and one external Integrity Annunciator light. The existing wiring and shielding, and Cirrus Antenna upgrade kit installation was inspected and determined to be IAW STC AML installation data.

A summary of the modification done to the aircraft is as follows:

1. New Cirrus Antenna Upgrade Kit and Antenna Cable was installed previously by Quest Aviation Inc. a Cirrus Service Center IAW Cirrus Service Bulletin SB 2X-34-23 R1 Revised 29 Jan. 2008.

2. Removed Garmin GNS 430 P/N: 011-00280-10 unit and installed Garmin GNS430W, P/N: 010-00412-40, S/N: 97110738, using the provision left behind from the standard 430 unit. Installation done IAW Garmin upgrade installation manual P/N: 190-00357-06 Rev B Dated January 3, 2007 and STC No. SA01933LA.

3. The GNS 430W was configured identical to the original 430 unit. Each interface was checked out IAW the 430W Installation Manual P/N: 190-00356-02 Section 5. A copy of the checkout log was completed and included with the aircrafts maintenance records.

4. Removed the Aircraft Flight Manual Supplement for the GNS 430 and installed a GNS 430 AFMS P/N: 190-00356-03 Rev A, FAA Approved date November 20, 2007 into the Aircraft Flight Manual.

5. Updated the aircraft Equipment List and Weight and Balance to reflect the new WAAS unit. The current electrical load analysis remains valid since the new unit draws the same of less current than the original unit.

Instructions for Continued Airworthiness (ICA)

1. GNS 430W - Included Garmin document P/N: 190-00356-65 Rev A, GNS 430W Instructions for Continued Airworthiness in the aircraft maintenance records.

Note: This supercedes ICAW data for the previously installed GNS 430.

END

☐ Additional Sheets Are Attached

FAA Airworthiness Directives Compliance Record

Company:

Category: Airframe

Manufacturer: Cirrus Design Corp.

Model: SR20

Position:

P/N:

S/N: 1258

Aircraft Registration No: N777ND

ATP Revision: 05/23/2018

Issue Number Effective Date Amendment #	Description	Complied	Method of Compliance	Recur	Next Due	Facility Cert No. / Type Authorized By Signed
2008-11-18 07/07/2008	To detect and correct leaks in the exhaust system, which could result in exhaust gases leaking into the cabin,contd.	05/16/2018 Hrs: 2553.2 C: --	C/W by Pressure check due each 100 hours	Yes	D: -- Hrs: 02/01/2653 C: --	/
2008-03-16 03/11/2008	To prevent the possibility of jamming of the rudder-aileron interconnect system, which may result in loss of,contd.	07/27/2007 Hrs: -- C: --	C/W 1/8/08 by complying with SB 2X-27-14 R3	No	D: -- Hrs: -- C: --	/ See A/C log books
2007-14-03 08/16/2007	To correct pick-up collar support fasteners of the CAPS, which could result in the premature separataion ,contd.	08/02/2007 Hrs: -- C: --	C/W 5/2/07 by complying with SB A2X95-10 R2 @	No	D: -- Hrs: -- C: --	/ See A/C log books
2006-21-03 11/17/2006	To detect, correct, & prevent overheating damage to the brake caliper piston O-ring seals, which could,contd.	04/28/2006 Hrs: -- C: --	C/W by completing S/B 2X-32-14 and installing kit P/N 70144-001 at 1209.0 ACTT.	No	D: -- Hrs: -- C: --	See A/C log books / See A/C log books
2006-19-10 10/24/2006	To prevent the crew seats from folding forward during emergency landing with dynamic loads with consequent,contd.	04/28/2008 Hrs: -- C: --	Part I C/W by installing SB 2X-25-17 R1 & adjusting pins IAW SB A2X-25-08. Part II C/W IAW SB 2X-25-06 R4 on 1-12-05 at 519.0, no further action needed.	No	D: -- Hrs: -- C: --	See A/C log books /

Issue Number Effective Date Amendment #	Description	Complied	Method of Compliance	Recur	Next Due	Facility Cert No. / Type Authorized By Signed
2006-07-06 05/11/2006	To detect, correct, & prevent damage to the fuel line and wire bundles, which could result in fuel leaks	08/18/2006 Hrs: -- C: --	C/W AD 2006-07-06 by installing fuel line shielding kit P/N 70133-001 IAW SB 2X-28-04 R1	No	D: -- Hrs: -- C: --	See A/C log books /
2005-17-19 C 10/13/2005	To prevent the crew seats from folding forward during emergency landing dynamic loads with consequent,contd.	04/28/2006 Hrs: 248-6 C: --	Verified AD 2005-17-19 is in compliance. Part I C/W by installing SB 2X-25-17 R1 & adjusting pins IAW SB A2X-25-08. Part II C/W IAW SB 2X-25-06 R4 on 1-12-05 at 519.0, no further action needed.	No	D: -- Hrs: -- C: --	See A/C log books /

Category: Engine **Position:** **ATP Revision:** 05/23/2018

Manufacturer: Teledyne Continental

Model: IO-360-ES **P/N:** IO-360-ES16B

S/N: 1000156

Issue Number Effective Date Amendment #	Description	Complied	Method of Compliance	Recur	Next Due	Facility Cert No. / Type Authorized By Signed
--		-- Hrs: -- C: --		--	D: -- Hrs: -- C: --	

Category: Propeller **Position:** **ATP Revision:** 05/23/2018

Manufacturer: Hartzell Propeller

Model: PHC-J3YF-1 **P/N:** Hartzell PHC-J3YF-1RF

S/N: FP1868B

Issue Number Effective Date Amendment #	Description	Complied	Method of Compliance	Recur	Next Due	Facility Cert No. / Type Authorized By Signed
--		-- Hrs: -- C: --		--	D: -- Hrs: -- C: --	

Category: Vacuum Pumps **Position:** **ATP Revision:** 05/23/2018

Manufacturer: Any Manufacturer
Model: Any Model

P/N: AA32[illegible]
S/N:

Issue Number Effective Date Amendment #	Description	Complied	Method of Compliance	Recur	Next Due	Facility Cert No. / Type Authorized By Signed
--		-- Hrs: -- C: --		--	D: -- Hrs: -- C: --	

Category: Fuel Pumps
Position:
ATP Revision: 05/23/2018
Manufacturer: Any Manufacturer
P/N: 5217-00-3
Model: Any Model
S/N: 2043

Issue Number Effective Date Amendment #	Description	Complied	Method of Compliance	Recur	Next Due	Facility Cert No. / Type Authorized By Signed
--		-- Hrs: -- C: --		--	D: -- Hrs: -- C: --	

Category: Global Positioning System
Position: 1
ATP Revision: 05/23/2018
Manufacturer: Garmin International
P/N: 011-01060-40
Model: GNS 430
S/N: 97110738

Issue Number Effective Date Amendment #	Description	Complied	Method of Compliance	Recur	Next Due	Facility Cert No. / Type Authorized By Signed
--		-- Hrs: -- C: --		--	D: -- Hrs: -- C: --	

Category: Magnetos
Position: LEFT
ATP Revision: 05/23/2018
Manufacturer: Teledyne Continental
P/N:
Model: S6-20 SERIES
S/N: D09CA083R

Issue Number Effective Date Amendment #	Description	Complied	Method of Compliance	Recur	Next Due	Facility Cert No. / Type Authorized By Signed
--		-- Hrs: -- C: --		--	D: -- Hrs: -- C: --	

Category: Governors
Position:
ATP Revision: 05/23/2018

Manufacturer: Any Manufacturer
P/N: D210760

Model: Any Model
S/N: 00131

Issue Number Effective Date Amendment #	Description	Complied	Method of Compliance	Recur	Next Due	Facility Cert No. / Type Authorized By Signed
2010-13-10 08/05/2010	To prevent loss of propeller pitch control, damage to the propeller governor, and internal damage to the,contd.	-- Hrs: -- C: --	N/A by S/N	No	D: -- Hrs: -- C: --	/
82-12-04 06/07/1982	COMBINED POWER TURBINE AND PROPELLER GOVERNOR ASSEMBLY	-- Hrs: -- C: --	N/A by Model	No	D: -- Hrs: -- C: --	/
81-25-01 12/07/1981	TO PREVENT THE POSSIBLE FAILURE OF THE FLYWEIGHTS	-- Hrs: -- C: --	N/A by Model	No	D: -- Hrs: -- C: --	/
75-12-07 06/06/1975	TO PREVENT THE POSSIBILITY OF LOSS OF PROPELLER PITCH CONTROL INCLUDING THE INABILITY TO FEATHER, CONTD.	-- Hrs: -- C: --	N/A by Model	No	D: -- Hrs: -- C: --	/
70-26-02 12/27/1970	TO PREVENT LOSS OF PROPELLER CONTROL	-- Hrs: -- C: --	N/A by Model	No	D: -- Hrs: -- C: --	/
70-16-01 08/04/1970	TO PREVENT UNCONTROLLED FEATHERING	-- Hrs: -- C: --	N/A by Model	No	D: -- Hrs: -- C: --	/

Issue Number Effective Date Amendment #	Description	Complied	Method of Compliance	Recur	Next Due	Facility Cert No. / Type Authorized By Signed
59-10-05 01/01/1959	PROVIDE A MEANS FOR FEATHERING THAT WILL BE INDEPENDENT OF THE PILOT VALVE INCORPORATED IN THE 5U21 GOVERNOR	-- Hrs: -- C: --	N/A by Model	No	D: -- Hrs: -- C: --	/
58-03-02 01/01/1958	TO PRECLUDE ADDITIONAL UNWANTED REVERSALS FROM THIS CAUSE, REMOVE THE SOLENOID VALVE ASSY FROM THE GOVERNOR	-- Hrs: -- C: --	N/A by Model	No	D: -- Hrs: -- C: --	/
58-18-01 01/01/1958	PROVIDE A MEANS FOR FEATHERING THAT WILL BE INDEPENDENT OF THE LOW PRESSURE RELIEF AND PILOT VALVES,CONTD.	-- Hrs: -- C: --	N/A by Model	No	D: -- Hrs: -- C: --	/
57-16-04 01/01/1957	TO PREVENT THE POSSIBLE OCCURRENCE OF PROPELLER REVERSAL RESULTING FROM OIL LEAKAGE CAUSED BY THE,CONTD.	-- Hrs: -- C: --	N/A by Model	No	D: -- Hrs: -- C: --	/
56-20-06 01/01/1956	SUBSEQUENT TO OCT. 15, 1956, NO HAMILTON STANDARD GOVERNOR DRIVE GEAR SHAFTS P/N 67035 SHALL BE USED IN THEIR,CONTD.	-- Hrs: -- C: --	N/A by Model	No	D: -- Hrs: -- C: --	/

Category: Global Positioning System
Manufacturer: Garmin International
Model: GNS 430

Position: 2
P/N: 011-01060-40
S/N: 97110592

ATP Revision: 05/23/2018

Issue Number Effective Date Amendment #	Description	Complied	Method of Compliance	Recur	Next Due	Facility Cert No. / Type Authorized By Signed
2001-23-17 12/28/2001	To prevent external noise from causing inaccurate course deviation displays in the GNS 430 unit's course,contd.	04/28/2006 Hrs: 248.6 C: --	D/N/A BY P/N AND S/N OF UNIT INSTALLED	No	D: -- Hrs: -- C: --	See A/C log books /

Category: Magnetos
Manufacturer: Teledyne Continental
Model: S6-20 SERIES

Position: RIGHT
P/N: BL-500556-3
S/N: D09CA085R

ATP Revision: 05/23/2018

Issue Number Effective Date Amendment #	Description	Complied	Method of Compliance	Recur	Next Due	Facility Cert No. / Type Authorized By Signed
--		-- Hrs: -- C: --		--	D: -- Hrs: -- C: --	

Category: Oil Coolers
Manufacturer: Any Manufacturer
Model: Any Model
Position:
P/N: 20455A
S/N: L08-6608-504
ATP Revision: 05/23/2018

Issue Number Effective Date Amendment #	Description	Complied	Method of Compliance	Recur	Next Due	Facility Cert No. / Type Authorized By Signed
--		-- Hrs: -- C: --		--	D: -- Hrs: -- C: --	

Category: Air Filter
Manufacturer: Brackett Aircraft
Model: Any Model
Position:
P/N:
S/N: BA-111
ATP Revision: 05/23/2018

Issue Number Effective Date Amendment #	Description	Complied	Method of Compliance	Recur	Next Due	Facility Cert No. / Type Authorized By Signed
84-26-02 01/29/1985	[Recurring] TO PREVENT POSSIBLE ENGINE POWER LOSS OR STOPPAGE CAUSED BY ENGINE INGESTION OF FRAGMENTS, CONTD.	-- Hrs: -- C: --	N/A effected parts are not installed.	Yes	D: -- Hrs: -- C: --	
80-08-14 04/22/1980	[Recurring] FOAM FILTER ELEMENT	-- Hrs: -- C: --	N/A by mfg date of aircraft	Yes	D: -- Hrs: -- C: --	

Category: Alternators
Manufacturer: Any Manufacturer
Model: No model selected
Position: 2
P/N: 653344
S/N: K-039998
ATP Revision: 05/23/2018

Issue Number Effective Date Amendment #	Description	Complied	Method of Compliance	Recur	Next Due	Facility Cert No. / Type Authorized By Signed
--		-- Hrs: -- C: --		--	D: -- Hrs: -- C: --	

Category: Alternators
Manufacturer: Hartzell Engine Tech
Model: Any Model

Position: 1
P/N: ES-4024LP
S/N: N021352

ATP Revision: 05/23/2018

Issue Number Effective Date Amendment #	Description	Complied	Method of Compliance	Recur	Next Due	Facility Cert No. / Type Authorized By Signed
76-02-07 02/02/1976	[Recurring] TO DETECT DEFECTIVE ALTERNATOR SLIP RING END BEARINGS AND MINIMIZE THE PROBABILITY OF IN-SERVICE FAILURES	-- Hrs: -- C: --		Yes	D: -- Hrs: -- C: --	
72-15-02 09/01/1972	[Recurring] TO PRECLUDE IN-SERVICE FAILURES OF ALTERNATOR COOLING FANS	-- Hrs: -- C: --		Yes	D: -- Hrs: -- C: --	
72-01-05 01/01/1972	Superseded by 72-15-02	-- Hrs: -- C: --		No	D: -- Hrs: -- C: --	

FAA Airworthiness Directives Compliance Record

Company:

Aircraft Registration No: N777ND

Category: Airframe

Position:

ATP Revision: 11/14/2016

Manufacturer: Cirrus Design Corp.

P/N:

Model: SR20

S/N: 1258

Issue Number Effective Date	Description	Complied	Method of Compliance	Recur?	Next Due	Cert No./Type Authorized Signed
2008-11-18 07/07/2008	To detect and correct leaks in the exhaust system, which could result in exhaust gases leaking into the cabin,contd.	11/08/2016 2480.0	C/W by Pressure check due each 100 hours	Y	2580.00	OW5R108N/CRS Mark Lehrkamp
2008-03-16 03/11/2008	To prevent the possibility of jamming of the rudder-aileron interconnect system, which may result in loss of,contd.	07/27/2007	C/W 1/8/08 by complying with SB 2X-27-14 R3	N		/ See A/C log books
2007-14-03 08/16/2007	To correct pick-up collar support fasteners of the CAPS, which could result in the premature separataion ,contd.	08/02/2007	C/W 5/2/07 by complying with SB A2X95-10 R2 @	Y		/ See A/C log books
2006-21-03 11/17/2006	To detect, correct, & prevent overheating damage to the brake caliper piston O-ring seals, which could,contd.	04/28/2006	C/W by completing S/B 2X-32-14 and installing kit P/N 70144-001 at 1209.0 ACTT.	N		/ See A/C log books
2006-19-10 10/24/2006	To prevent the crew seats from folding forward during emergency landing with dynamic loads with consequent,contd.	04/28/2008	Part I C/W by installing SB 2X-25-17 R1 & adjusting pins IAW SB A2X-25-08. Part II C/W IAW SB 2X-25-06 R4 on 1-12-05 at 519.0, no further action needed.	N		/
2006-07-06 05/11/2006	To detect, correct, & prevent damage to the fuel line and wire bundles, which could result in fuel leaks	08/18/2006	C/W AD 2006-07-06 by installing fuel line shielding kit P/N 70133-001 IAW SB 2X-28-04 R1	N		/

Issue Number Effective Date	Description	Complied	Method of Compliance	Recur?	Next Due	Cert No./Type Authorized Signed
2005-17-19 C 10/13/2005	To prevent the crew seats from folding forward during emergency landing dynamic loads with consequent,contd.	04/28/2006 248-6	Verified AD 2005-17-19 is in compliance. Part I C/W by installing SB 2X-25-17 R1 & adjusting pins IAW SB A2X-25-08. Part II C/W IAW SB 2X-25-06 R4 on 1-12-05 at 519.0, no further action needed.	N		/

Category: Engine
Manufacturer: Teledyne Continental
Model: IO-360-ES
Position:
P/N: IO-360-ES16B
S/N: 1000156
ATP Revision: 11/14/2016

Issue Number Effective Date	Description	Complied	Method of Compliance	Recur?	Next Due	Cert No./Type Authorized Signed
						/

Category: Propeller
Manufacturer: Hartzell Propeller
Model: PHC-J3YF-1
Position:
P/N: Hartzell PHC-J3YF-1RF
S/N: FP1868B
ATP Revision: 11/14/2016

Issue Number Effective Date	Description	Complied	Method of Compliance	Recur?	Next Due	Cert No./Type Authorized Signed
						/

Category: Magnetos
Manufacturer: Slick
Model: 6314
Position:
P/N:
S/N: D09CA083R
ATP Revision: 11/14/2016

Issue Number Effective Date	Description	Complied	Method of Compliance	Recur?	Next Due	Cert No./Type Authorized Signed
						/

Category: Magnetos
Manufacturer: Slick
Model: 6314
Position:
P/N:
S/N:
ATP Revision: 11/14/2016

Issue Number Effective Date	Description	Complied	Method of Compliance	Recur?	Next Due	Cert No./Type Authorized Signed

sue Number ffective Date	Description	Complied	Method of Compliance	Recur?	Next Due	Cert No./Type Authorized Signed

Category: Vacuum Pumps
Manufacturer: Any Manufacturer
Model: Any Model
Position:
P/N: AA3215CC
S/N:
ATP Revision: 11/14/2016

Issue Number Effective Date	Description	Complied	Method of Compliance	Recur?	Next Due	Cert No./Type Authorized Signed

Category: Fuel Pumps
Manufacturer: Any Manufacturer
Model: Any Model
Position:
P/N: 5217-00-3
S/N: 2043
ATP Revision: 11/14/2016

Issue Number Effective Date	Description	Complied	Method of Compliance	Recur?	Next Due	Cert No./Type Authorized Signed

Category: Global Positioning System
Manufacturer: Garmin International
Model: GNS 430
Position:
P/N:
S/N:
ATP Revision: 11/14/2016

Issue Number Effective Date	Description	Complied	Method of Compliance	Recur?	Next Due	Cert No./Type Authorized Signed

Category: Magnetos
Manufacturer: Teledyne Continental
Model: S6-20 SERIES
Position: LEFT
P/N:
S/N: D09CA083R
ATP Revision: 11/14/2016

Issue Number Effective Date	Description	Complied	Method of Compliance	Recur?	Next Due	Cert No./Type Authorized Signed

tegory: Governors
Manufacturer: Any Manufacturer
Position:
P/N: D210760
ATP Revision: 11/14/2016

Model: Any Model **S/N:** 00131

Issue Number Effective Date	Description	Complied	Method of Compliance	Recur?	Next Due	Cert No./Type Authorized Signed
2010-13-10 08/05/2010	To prevent loss of propeller pitch control, damage to the propeller governor, and internal damage to the.contd.			N		
82-12-04 06/07/1982	COMBINED POWER TURBINE AND PROPELLER GOVERNOR ASSEMBLY		N/A by Model	N		
81-25-01 12/07/1981	TO PREVENT THE POSSIBLE FAILURE OF THE FLYWEIGHTS		N/A by Model	N		
75-12-07 06/06/1975	TO PREVENT THE POSSIBILITY OF LOSS OF PROPELLER PITCH CONTROL INCLUDING THE INABILITY TO FEATHER, CONTD.		N/A by Model	N		
70-26-02 12/27/1970	TO PREVENT LOSS OF PROPELLER CONTROL		N/A by Model	N		
70-16-01 08/04/1970	TO PREVENT UNCONTROLLED FEATHERING		N/A by Model	N		
59-10-05 01/01/1959	PROVIDE A MEANS FOR FEATHERING THAT WILL BE INDEPENDENT OF THE PILOT VALVE INCORPORATED IN THE 5U21 GOVERNOR		N/A by Model	N		
58-03-02 01/01/1958	TO PRECLUDE ADDITIONAL UNWANTED REVERSALS FROM THIS CAUSE, REMOVE THE SOLENOID VALVE ASSY FROM THE GOVERNOR		N/A by Model	N		
58-18-01 01/01/1958	PROVIDE A MEANS FOR FEATHERING THAT WILL BE INDEPENDENT OF THE LOW PRESSURE RELIEF AND PILOT VALVES,CONTD.		N/A by Model	N		

ue Number ffective Date	Description	Complied	Method of Compliance	Recur?	Next Due	Cert No./Type Authorized Signed
57-16-04 /01/1957	TO PREVENT THE POSSIBLE OCCURRENCE OF PROPELLER REVERSAL RESULTING FROM OIL LEAKAGE CAUSED BY THE,CONTD.		N/A by Model	N		/

56-20-06 01/01/1956	SUBSEQUENT TO OCT. 15, 1956, NO HAMILTON STANDARD GOVERNOR DRIVE GEAR SHAFTS P/N 67035 SHALL BE USED IN THEIR,CONTD.		N/A by Model	N		/

Category: Global Positioning System **Position:** **ATP Revision:** 11/14/2016

Manufacturer: Garmin International **P/N:**

Model: GNS 430 **S/N:**

Issue Number Effective Date	Description	Complied	Method of Compliance	Recur?	Next Due	Cert No./Type Authorized Signed
2001-23-17 12/28/2001	To prevent external noise from causing inaccurate course deviation displays in the GNS 430 unit's course,contd.	04/28/2006 248.6	D/N/A BY P/N AND S/N OF UNIT INSTALLED	N		/

tegory: Magnetos **Position:** RIGHT **ATP Revision:** 11/14/2016

Manufacturer: Teledyne Continental **P/N:** BL-500556-3

Model: S6-20 SERIES **S/N:** D09CA085R

Issue Number Effective Date	Description	Complied	Method of Compliance	Recur?	Next Due	Cert No./Type Authorized Signed
						/

Category: Oil Coolers **Position:** **ATP Revision:** 11/14/2016

Manufacturer: Any Manufacturer **P/N:** 20455A

Model: Any Model **S/N:** L08-6608-504

Issue Number Effective Date	Description	Complied	Method of Compliance	Recur?	Next Due	Cert No./Type Authorized Signed
						/

Category: Air Filter **Position:** **ATP Revision:** 11/14/2016

Manufacturer: Brackett Aircraft **P/N:**

odel: Any Model **S/N:** BA-111

Issue Number Effective Date	Description	Complied	Method of Compliance	Recur?	Next Due	Cert No./Type Authorized Signed

Category: Alternators
Manufacturer: Any Manufacturer
Model: No model selected
Position: 1
P/N: ES-4024LP
S/N: J020326
ATP Revision: 11/14/2016

Issue Number Effective Date	Description	Complied	Method of Compliance	Recur?	Next Due	Cert No./Type Authorized Signed

Category: Magnetos
Manufacturer: Bendix Corporation
Model: S-20 SERIES
Position:
P/N:
S/N:
ATP Revision: 11/14/2016

Issue Number Effective Date	Description	Complied	Method of Compliance	Recur?	Next Due	Cert No./Type Authorized Signed
2005-12-06 07/19/2005	To prevent failure of the magneto impulse coupling assembly and possible engine failure			Y		
96-12-07 07/18/1996	Superseded by 2005-12-06			Y		
94-01-03 R2 06/28/1995	TO PREVENT MAGNETO FAILURE AND SUBSEQUENT ENGINE FAILURE			N		
94-06-09 05/20/1994	TO PREVENT INJURY OR DEATH TO GROUND PERSONNEL DUE TO A NON-GROUNDED MAGNETO			N		
82-20-01 06/14/1983	TO PREVENT FAILURE OF IMPULSE COUPLING DUE TO IMPROPERLY HEAT TREATED (SOFT) FLYWEIGHTS RESULTING IN ENGINE,CONTD.			N		
74-26-09 12/24/1974	S-20,-200,-1200 SERIES MAGNETOS			N		

sue Number ffective Date	Description	Complied	Method of Compliance	Recur?	Next Due	Cert No./Type Authorized Signed
73-07-04 /11/1973	Superseded by 94-01-03			N		/

Category: Airframe
Manufacturer: Cirrus Design Corp. **Model:** SR20 **P/N:** **S/N:** 1258 8/2/2013

FAA AD # Eff Date	Description	Complied Date & Time	Amendment Number Method of Compliance / Applicability	One Time	Recurring	Next Due	Authorized Signature & Number
2008-03-16 3/11/2008	To prevent the possibility of jamming of the rudder-aileron interconnect system, which may result in loss of,contd.	7/27/2007	C/W 1/8/08 by complying with SB 2X-27-14 R3	X			See A/C log books
2006-19-10 10/24/2006	To prevent the crew seats from folding forward during emergency landing with dynamic loads with consequent,contd.	4/28/2008	Part I C/W by installing SB 2X-25-17 R1 & adjusting pins IAW SB A2X-25-08. Part II C/W IAW SB 2X-25-06 R4 on 1-12-05 at 519 0, no further action needed.	X			
2005-17-19 C 10/13/2005	To prevent the crew seats from folding forward during emergency landing dynamic loads with consequent,contd.	4/28/2006 248-6	Verified AD 2005-17-19 is in compliance. Part I C/W by installing SB 2X-25-17 R1 & adjusting pins IAW SB A2X-25-08. Part II C/W IAW SB 2X-25-06 R4 on 1-12-05 at 519 0, no further action needed.	X			
2002-21-02 11/8/2002	To prevent loss of the self- locking retaining nut on the roll and yaw trim cartridges during flight, which,contd.	4/28/2006 248.6	D/N/A BY S/N (1005-1241)	X			
2002-24-08 1/24/2003	To prevent failure of the Cirrus Airplane Parachute System (CAPS) activation system in an emergency,contd.	1/2/2003 35.6	D/N/A BY S/N (1005-1195)	X			
2008-14-13 8/14/2008	To prevent in-flight failure of the cabin door, which could result in door separation from the airplane		N/A by serial number	X			
2006-21-03 11/17/2006	To detect, correct, & prevent overheating damage to the brake caliper piston O-ring seals, which could,contd.	4/28/2006	C/W by completing S/B 2X-32-14 and installing kit P/N 70144-001 at 1209.0 ACTT.	X			See A/C log books
2006-07-06 5/11/2006	To detect, correct, & prevent damage to the fuel line and wire bundles, which could result in fuel leaks	8/18/2006	C/W AD 2006-07-06 by installing fuel line shielding kit P/N 70133-001 IAW SB 2X-28-04 R1	X			
2001-25-03 12/17/2001	To detect and replace understrength rivets in the elevator and rudder, which could result in failure,contd.	4/28/2006 248.6	D/N/A BY A/C S/N (1134-1159)	X			
2008-11-18 7/7/2008	To detect and correct leaks in the exhaust system, which could result in exhaust gases leaking into the cabin,contd.	8/28/2013 Hobbs: 1206.6	C/W by Pressure check due each 100 hours		X	Next Due Hobl	OW5R108N CRS Mark Lehrkamp
2007-14-03 8/16/2007	To correct pick-up collar support fasteners of the CAPS, which could result in the premature separataion ,contd.	8/2/2007	C/W 5/2/07 by complying with SB A2X95-10 R2 @	X			See A/C log books
2002-05-05 3/19/2002	Superseded by 2002-24-08			X			

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Category: Engine
Manufacturer: Teledyne Continental **Model:** IO-360-ES **P/N:** IO-360-ES16B **S/N:** 1000156 8/2/2013

FAA AD # Eff Date	Description	Complied Date & Time	Amendment Number Method of Compliance / Applicability	One Time	Recurring	Next Due	Authorized Signature & Number
88-03-06 4/15/1988	TO PREVENT POSSIBLE LOSS OF ENGINE OIL AND SUBSEQUENT ENGINE FAILURE	4/28/2006 248.6	D/N/A ENGINE NOT EQUIPPED WITH TCM OIL FILTER	X			

Category: Engine
Manufacturer: Teledyne Continental **Model:** IO-360-ES **P/N:** IO-360-ES16B **S/N:** 1000156 8/2/2013

FAA AD # Eff Date	Description	Complied Date & Time	Amendment Number Method of Compliance / Applicability	One Time	Recurring	Next Due	Authorized Signature & Number
91-19-03 9/29/1991	TO PREVENT OPERATION WITH COLLAPSED OIL FILTER ELEMENTS, WHICH CAN RESULT IN LOSS OF OIL PRESSURE, CONTD	4/28/2006 248.6	D/N/A OIL FILTER DATE CODE IS OK	X			
2009-24-52 E 11/18/2009	To prevent excessive hydraulic lifter wear, which can result in loss of engine power & loss of control of the airplane	6/2/2010 828.7 Hobbs	N/A by date of manufacturer	X			
2010-11-04 6/16/2010	To prevent excessive hydraulic lifter wear, which can result in loss of engine power & loss of control of the airplane		N/A by build date of the engine	X			
92-04-09 6/22/1993	TO PREVENT ENGINE FAILURE CAUSED BY FAILURE OF THE ROCKER SHAFT HOLD DOWN STUD	4/28/2006 248.6	D/N/A BY ENGINE S/N LISTED IN SB M92-4 R1	X			
2002-13-04 C 7/12/2002	To prevent engine failure and loss of control of the airplane due to migration of the magneto impulse,contd	4/28/2006 248.6	D/N/A BY MAGNETO S/N INSTALLED	X			
2011-26-07 1/24/2012	To prevent engine failure and loss of control of the airplane due to migration of the magneto impulse,contd.	8/8/2012 Hobbs: 1091.3	N/A BY Model Installed	X			
97-26-17 C 1/23/1998	TO PREVENT CRANKSHAFT FAILURE AND SUBSEQUENT ENGINE FAILURE	4/28/2006 248.6	D/N/A BY MFG DATE OF ENGINE		X		
2000-23-21 12/12/2000	To prevent crankshaft connecting rod journal fracture, which could result in total engine power,contd.	4/28/2006 248.6	D/N/A BY MFG DATE OF ENGINE	X			
2012-03-06 C 2/24/2012	To prevent an in-flight engine shutdown due to a failed fuel servo diaphragm, and damage to the airplane	8/8/2012 Hobbs: 1091.3	D/N/A Affected Part Not Installed	X			
93-10-02 8/12/1993	TO PREVENT AN ENGINE FAILURE DUE TO A MISSING CYLINDER VALVE RETAINER KEY	4/28/2006 248.6	D/N/A BY MFG DATE OF ENGINE	X			
92-02-20 E R1 2/3/1992	Superseded by 92-04-09			X			
2009-24-51 E 11/16/2009	Superseded by 2009-24-52			X			
2000-11-51 E 6/7/2000	Superseded by 2002-13-04			X			
2000-08-51 E 4/28/2000	Superseded by 2000-23-21			X			

Category: Propeller
Manufacturer: Hartzell Propeller **Model:** PHC-J3YF-1 **P/N:** Hartzell PHC-J3YF-1RI **S/N:** FP1868B 8/2/2013

FAA AD # Eff Date	Description	Complied Date & Time	Amendment Number Method of Compliance / Applicability	One Time	Recurring	Next Due	Authorized Signature & Number
2001-07-03 C 6/4/2001	To prevent propeller failure of the propellers returned to service by BASCO, & possible loss of airplane control	4/28/2006 248.6	D/N/A BY PROPELLER S/N INSTALLED	X			
94-17-13 9/15/1994	TO PREVENT POSSIBLE PROPELLER HUB FAILURE DUE TO CRACKS THAT ORIGINATE IN THE GREASE FITTING HOLES ON THE,CONTD	4/28/2006 248.6	D/N/A BY PROPELLER S/N INSTALLED		X		
2002-09-08 6/13/2002	To prevent failure of the propeller blade from fatigue cracks in the blade shank radius, which can,contd.	4/28/2006 248.6	D/N/A BY PROPELLER S/N INSTALLED	X			
2007-26-09 1/30/2008	To prevent failure of the propeller blade from fatigue cracks in the aluminum blade shank radius, which can,contd		D/N/A BY PROPELLER S/N INSTALLED	X			
2005-14-11 8/17/2005	To prevent blade failure that could result in separation of a propeller blade and loss of control of the airplane	4/28/2006 248.6	D/N/A PROPELLER NOT SERVICED AT SOUTHERN CALIFORNIA PROPELLER SERVICE	X			
93-16-14 1/5/1994	Superseded by 94-17-13				X		
89-22-05 L 11/16/1989	Superseded by 93-16-14				X		
70-16-03 R 1/1/1970	Superseded by 77-12-06			X			
73-10-03 1/1/1973	Superseded by 77-12-06			X			
74-15-02 1/1/1974	Superseded by 77-12-06			X			
75-07-05 5/1/1977	Superseded by 77-12-06			X			
70-02-01 1/1/1970	Superseded by 73-10-03			X			
77-12-06 R(2) 12/21/1977	Superseded by 2002-09-08				X		

Category: Vacuum Pumps
Manufacturer: Any Manufacturer **Model:** Any Model **P/N:** AA3215CC **S/N:** 8/2/2013

FAA AD # Eff Date	Description	Complied Date & Time	Amendment Number Method of Compliance / Applicability	One Time	Recurring	Next Due	Authorized Signature & Number

Category: Vacuum Pumps
Manufacturer: Any Manufacturer **Model:** Any Model **P/N:** AA3215CC **S/N:** 8/2/2013

FAA AD # Eff. Date	Description	Complied Date & Time	Amendment Number Method of Compliance / Applicability	One Time	Recurring	Next Due	Authorized Signature & Number
64-01-01 1/9/1964	TO REMOVE DEFECTIVE SPLINED DRIVE COUPLINGS WHICH RENDER THE VACUUM FLIGHT INSTRUMENTS INOPERATIVE	4/28/2006 248.6	D/N/A TO VACUUM PUMP INSTALLED	X			
2006-03-08 3/10/2006	To prevent the vacuum pump failure or malfunction during instrument flight rules (IFR) flight that could lead,contd.		D/N/A to vacuum pumps installed	X			
86-01-06 1/31/1986	TO PREVENT PREMATURE FAILURE OF THE AIRBORNE DRY AIR AND AUXILIARY DRY AIR PUMPS	4/28/2006 248.6	D/N/A TO VACUUM PUMP INSTALLED	X			
76-16-02 8/11/1976	TO PREVENT LOSS OF VACUUM PRESSURE WITH THE ACCOMPANYING LOSS OF DIRECTIONAL OR ATTITUDE GYRO FUNCTION	4/28/2006 248.6	D/N/A TO VACUUM PUMP INSTALLED	X			
98-23-01 11/20/1998	TO PREVENT FAILURE OF THE PRIMARY DRY AIR PUMP CAUSED BY DEFECTIVE FLEXIBLE COUPLING, WHICH COULD,CONTD	4/28/2006 248.6	D/N/A TO VACUUM PUMP INSTALLED	X			
79-13-08 6/7/1979	TO PREVENT CATASTROPHIC FAILURE OF THE PUMP AND SUBSEQUENT LOSS OF THE VACUUM SYSTEM	4/28/2006 248.6	D/N/A TO VACUUM PUMP INSTALLED	X			

Category: GPS/NAV/COM
Manufacturer: Garmin International **Model:** GNS 430 **P/N:** **S/N:** 8/2/2013

FAA AD # Eff. Date	Description	Complied Date & Time	Amendment Number Method of Compliance / Applicability	One Time	Recurring	Next Due	Authorized Signature & Number
2001-23-17 12/28/2001	To prevent external noise from causing inaccurate course deviation displays in the GNS 430 unit's course,contd.	4/28/2006 248.6	D/N/A BY P/N AND S/N OF UNIT INSTALLED	X			

Category: Fuel Pumps
Manufacturer: Any Manufacturer **Model:** Any Model **P/N:** 5217-00-3 **S/N:** 2043 8/2/2013

FAA AD # Eff. Date	Description	Complied Date & Time	Amendment Number Method of Compliance / Applicability	One Time	Recurring	Next Due	Authorized Signature & Number
55-26-02 8/1/1956	TO PREVENT TF-1100 FUEL PUMP FAILURE RESULTING FROM EXCESSIVE DRIVE PIN WEAR		N/A by Model		X		
54-21-01 11/30/1954	TO PREVENT LOOSENING OF THE SEDIMENT BOWL AND THE SURGE CHAMBER		N/A by Model	X			
93-11-11 6/21/1993	TO PREVENT DISRUPTION OF FUEL FLOW TO THE ENGINE, WHICH CAN RESULT IN A LOSS OF ENGINE POWER		N/A by Model	X			
93-05-21 L 3/25/1993	Superseded by 93-11-11			X			

Category: Fuel Pumps
Manufacturer: Any Manufacturer **Model:** Any Model **P/N:** 5217-00-3 **S/N:** 2043 8/2/2013

FAA AD # Eff Date	Description	Complied Date & Time	Amendment Number Method of Compliance / Applicability	One Time	Recurring	Next Due	Authorized Signature & Number
92-20-07 L 10/5/1992	Superseded by 93-05-21			X			
55-19-02 1/1/1955	Superseded by 55-26-02			X			
00-00-07 10/10/2001	Important notice for Chandler Evans engine fuel pumps listed in Honeywell International, Inc AD 2001-17-15		N/A by Model	X			

Category: Governors
Manufacturer: Any Manufacturer **Model:** Any Model **P/N:** D210760 **S/N:** 00131 8/2/2013

FAA AD # Eff Date	Description	Complied Date & Time	Amendment Number Method of Compliance / Applicability	One Time	Recurring	Next Due	Authorized Signature & Number
70-16-01 8/4/1970	TO PREVENT UNCONTROLLED FEATHERING		N/A by Model	X			
57-16-04 1/1/1957	TO PREVENT THE POSSIBLE OCCURRENCE OF PROPELLER REVERSAL RESULTING FROM OIL LEAKAGE CAUSED BY THE,CONTD.		N/A by Model	X			
81-25-01 12/7/1981	TO PREVENT THE POSSIBLE FAILURE OF THE FLYWEIGHTS		N/A by Model	X			
75-12-07 6/6/1975	TO PREVENT THE POSSIBILITY OF LOSS OF PROPELLER PITCH CONTROL INCLUDING THE INABILITY TO FEATHER. CONTD.		N/A by Model	X			
2010-13-10 8/5/2010	To prevent loss of propeller pitch control, damage to the propeller governor, and internal damage to the,contd.			X			
70-26-02 12/27/1970	TO PREVENT LOSS OF PROPELLER CONTROL		N/A by Model	X			
58-03-02 1/1/1958	TO PRECLUDE ADDITIONAL UNWANTED REVERSALS FROM THIS CAUSE. REMOVE THE SOLENOID VALVE ASSY FROM THE GOVERNOR		N/A by Model	X			
56-02-02 1/1/1956	Superseded by 56-20-06			X			
56-20-06 1/1/1956	SUBSEQUENT TO OCT. 15, 1956, NO HAMILTON STANDARD GOVERNOR DRIVE GEAR SHAFTS P/N 67035 SHALL BE USED IN THEIR,CONTD.		N/A by Model	X			
59-10-05 1/1/1959	PROVIDE A MEANS FOR FEATHERING THAT WILL BE INDEPENDENT OF THE PILOT VALVE INCORPORATED IN THE 5U21 GOVERNOR		N/A by Model	X			

Category: Governors
Manufacturer: Any Manufacturer **Model:** Any Model **P/N:** D210760 **S/N:** 00131 8/2/2013

FAA AD # Eff Date	Description	Complied Date & Time	Amendment Number Method of Compliance / Applicability	One Time	Recurring	Next Due	Authorized Signature & Number
58-18-01 1/1/1958	PROVIDE A MEANS FOR FEATHERING THAT WILL BE INDEPENDENT OF THE LOW PRESSURE RELIEF AND PILOT VALVES,CONTD		N/A by Model	X			
82-12-04 6/7/1982	COMBINED POWER TURBINE AND PROPELLER GOVERNOR ASSEMBLY		N/A by Model	X			

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Category: Oil Coolers
Manufacturer: Any Manufacturer **Model:** Any Model **P/N:** 20455A **S/N:** L08-6608-504 8/2/2013

FAA AD # Eff Date	Description	Complied Date & Time	Amendment Number Method of Compliance / Applicability	One Time	Recurring	Next Due	Authorized Signature & Number
80-25-07 R1 9/24/1981	TO PREVENT THE LOSS OF ENGINE OIL		N/A by Model		X		

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Category: Air Filter
Manufacturer: Brackett Aircraft **Model:** Any Model **P/N:** **S/N:** BA-111 8/2/2013

FAA AD # Eff Date	Description	Complied Date & Time	Amendment Number Method of Compliance / Applicability	One Time	Recurring	Next Due	Authorized Signature & Number
81-15-03 7/20/1981	TO PREVENT POSSIBLE FAILURE OF THE ALUMINUM AIR FILTER RETAINER SCREEN OR GASKETS WITH POTENTIAL. CONTD		N/A by Model	X			
96-09-06 6/7/1996	TO PREVENT GASKET PARTICLES FROM ENTERING THE CARBURETOR BECAUSE OF AIR FILTER GASKET FAILURE, WHICH COULD,CONTD.		N/A by Model		X		
2002-26-03 2/18/2003	To detect & correct incorrect installation of the air filter, which could result in failure of the air filter		N/A by Model	X			
95-03-02 3/17/1995	Superseded by 96-09-06				X		
78-25-05 1/1/1978	Superseded by 81-15-03			X			

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Category: Alternators
Manufacturer: Any Manufacturer **Model:** **P/N:** ES-4024LP **S/N:** J020326 8/2/2013

FAA AD # Eff Date	Description	Complied Date & Time	Amendment Number Method of Compliance / Applicability	One Time	Recurring	Next Due	Authorized Signature & Number
72-15-02 9/1/1972	TO PRECLUDE IN-SERVICE FAILURES OF ALTERNATOR COOLING FANS		N/A by Model		X		

Category: Alternators
Manufacturer: Any Manufacturer Model: P/N: ES-4024LP S/N: J020328 8/2/2013

FAA AD # Eff Date	Description	Complied Date & Time	Amendment Number Method of Compliance / Applicability	One Time	Recurring	Next Due	Authorized Signature & Number
76-02-07 2/2/1976	TO DETECT DEFECTIVE ALTERNATOR SLIP RING END BEARINGS AND MINIMIZE THE PROBABILITY OF IN SERVICE FAILURES		N/A by Model		X		
72-01-05 1/1/1972	Superseded by 72-15-02			X			